

WHAT IS CLAIMED IS:

1. A method for fabricating a connector for a gas turbine engine, the connector sized to receive a threaded apparatus, the connector including a first side, a second side, and an opening extending therebetween and defined by an inner surface, said method comprising the steps of:

5 forming threads into the connector inner surface; and

 forming a cutback to extend into the connector inner surface to reduce stresses induced into the plurality of threads by the threaded apparatus.

2. A method in accordance with Claim 1 wherein the plurality of threads include a first thread portion which is the closest thread to the connector first side, said step of forming a cutback further comprises the step of extending the cutback through the first thread portion.

3. A method in accordance with Claim 2 wherein the connector opening has a center axis of symmetry extending between the connector top and bottom sides, each of the threads includes a root and a crest, said step of extending the cutback further comprises the step of extending the cutback through the first thread portion crest.

4. A method in accordance with Claim 3 wherein the connector cutback includes a first end, and a second end between the first thread portion and the connector second side, said step of extending the cutback further comprises the step of extending the cutback obliquely through the first thread portion with respect to the connector opening center axis of symmetry.

5. A method in accordance with Claim 4 wherein said step of extending the cutback obliquely further comprises the step of extending the cutback through the first thread portion such that the cutback first end is a first distance from the connector opening center axis of symmetry that is longer than a second distance that the cutback second end is located relative to the same center axis of symmetry.

6. A receptacle for receiving a threaded apparatus, said receptacle comprising a top side, a bottom side, and an opening extending therethrough, said opening defined by a receptacle inner surface, said inner surface comprising a plurality of threads extending radially outward from said inner surface into said opening, said plurality of threads comprising a cutback and a first thread portion, said cutback configured to reduce a stiffness factor of said first thread portion.

7. A receptacle in accordance with Claim 6 wherein said inner surface first thread portion adjacent said receptacle opening top side.

8. A receptacle in accordance with Claim 6 wherein said receptacle opening has a center axis of symmetry extending from said top side to said bottom side, each said thread comprises a root and a crest, said cutback extending through at least one of said thread crests.

9. A receptacle in accordance with Claim 8 wherein said cutback extends through said first thread portion crest.

10. A receptacle in accordance with Claim 8 wherein said cutback oblique with respect to said center axis of symmetry.

11. A receptacle in accordance with Claim 8 wherein said cutback comprises a first end and a second end, said first end extending through said first thread portion, said second end between said receptacle first thread portion and said receptacle bottom side, said cutback first end a first distance from said receptacle center axis of symmetry, said cutback second end a second distance from said receptacle axis of symmetry.

12. A receptacle in accordance with Claim 11 wherein said cutback first distance longer than said cutback second distance.

13. A gas turbine engine comprising a thrust link comprising a first connector, a second connector, and a span section extending therebetween, each said thrust link connector comprising a first side, a second side, and an opening extending

therebetween and defined by an inner surface, said inner surface comprising a plurality of threads extending radially outward into said connector opening, said plurality of threads comprising a cutback configured to reduce stresses induced into said plurality of threads.

5 14. A gas turbine engine in accordance with Claim 13 wherein each said thrust link connector opening has a center axis of symmetry extending from said connector first side to said connector second side, said cutback oblique with respect to said center axis of symmetry.

10 15. A gas turbine engine in accordance with Claim 14 wherein said plurality of threads comprise a first thread portion, each said thread comprises a crest and a root, said cutback extending through said first thread portion crest.

16. A gas turbine engine in accordance with Claim 15 wherein said thrust link connector first thread portion adjacent at least one of said connector first side and said connector second side.

15 17. A gas turbine engine in accordance with Claim 15 wherein said thrust link connector cutback comprises a first end and a second end, said cutback first end extending through said thrust link connector first thread portion, said cutback second end between said connector first thread portion and said thrust link connector second sides.

20 18. A gas turbine engine in accordance with Claim 17 wherein said thrust link connector cutback first end a first distance from said thrust link connector opening center axis of symmetry, said thrust link connector cutback second end a second distance from said thrust link connector opening center axis of symmetry.

25 19. A gas turbine engine in accordance with Claim 18 wherein said thrust link connector cutback first distance longer than said thrust link connector cutback second distance.

20. A gas turbine engine in accordance with Claim 13 wherein said thrust link connector cutback further configured to reduce a stiffness factor of at least a portion of said plurality of threads.